

## Development of High-fidelity Re-entry Safety Analysis Methods

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### ● Abstract

High-fidelity re-entry safety analysis code LS-DARC is under the development in JAXA in order to realize the accurate EC predictions and the design-for-demise for the rocket upper stages and space-crafts. 6 Dof trajectory analysis, aerodynamics, heat flux, 3D thermal conductance can be considered by the coupling analysis code based on the reduced order models. Since models in LS-DARC is versatile and its turn-around time is very short, it has been applied wide range of aerospace engineering applications such as the research and development activities for Japan's flagship launch vehicles, satellites, spacecrafts, and re-usable space transportation systems.

### ● Reasons and benefits of using JAXA Supercomputer System

Probabilistic re-entry safety analysis based on the multi disciplinary coupling analysis with considering various uncertainty factors is essential. In addition consideration on the design parameter effect should also be considered. In addition, in-house computational environment is essential also for security reason. Consequently, JSS2 is essential.

### ● Achievements of the Year

In FY2020, the high-fidelity re-entry safety analysis code "LS-DARC" was validated against simple geometries presented in open literatures, experimental data acquired by the high enthalpy wind tunnel "HIEST", and flight data acquired by the re-entry data measurement system installed in H-IIB launch vehicle in terms of aerodynamic force and wall heat flux prediction. The computed results of those validation cases showed reasonable agreement with computed results by other CFD or similar analysis tools, experimental data, and flight data. Figure 1 shows pressure coefficient  $C_p$  and wall heat flux distributions on the appollo capsule geometry. The  $C_p$  and wall heat flux profiles computed by LS-DARC compare well to the flight data. As a result, the capability of LS-DARC was

confirmed.

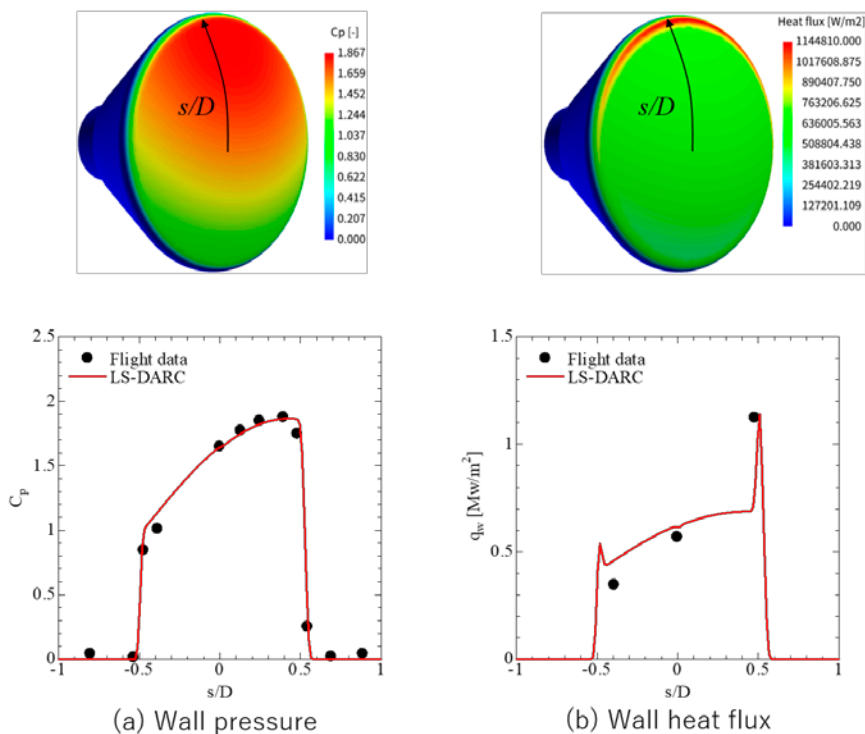


Fig. 1: Comparison of wall pressure  $C_p$  and heat flux distributions on the Apollo capsule

● Publications

N/A

● Usage of JSS

● Computational Information

Process Parallelization Methods	MPI
Thread Parallelization Methods	MPI
Number of Processes	32 - 192
Elapsed Time per Case	168 Hour(s)

- **Resources Used(JSS2)**

Fraction of Usage in Total Resources\*1(%): 0.03

Details

Computational Resources		
System Name	Amount of Core Time (core x hours)	Fraction of Usage*2(%)
SORA-MA	21,367.69	0.00
SORA-PP	1,732.79	0.01
SORA-LM	0.00	0.00
SORA-TPP	11,463.61	1.08

File System Resources		
File System Name	Storage Assigned (GiB)	Fraction of Usage*2(%)
/home	591.04	0.54
/data	18,763.54	0.36
/tmp	6,434.75	0.55

Archiver Resources		
Archiver Name	Storage Used (TiB)	Fraction of Usage*2(%)
J-SPACE	4.46	0.15

\*1: Fraction of Usage in Total Resources: Weighted average of three resource types (Computing, File System, and Archiver).

\*2: Fraction of Usage : Percentage of usage relative to each resource used in one year.

- **Resources Used(JSS3)**

Fraction of Usage in Total Resources\*1(%): 0.10

Details

Computational Resources		
System Name	Amount of Core Time (core x hours)	Fraction of Usage*2(%)
TOKI-SORA	2,822.43	0.00
TOKI-RURI	14,033.04	0.08
TOKI-TRURI	106,549.08	8.59

File System Resources		
File System Name	Storage Assigned (GiB)	Fraction of Usage*2(%)
/home	844.77	0.58
/data	82,974.73	1.39
/ssd	7,279.58	3.80

Archiver Resources		
Archiver Name	Storage Used (TiB)	Fraction of Usage*2(%)
J-SPACE	4.46	0.15

\*1: Fraction of Usage in Total Resources: Weighted average of three resource types (Computing, File System, and Archiver).

\*2: Fraction of Usage : Percentage of usage relative to each resource used in one year.